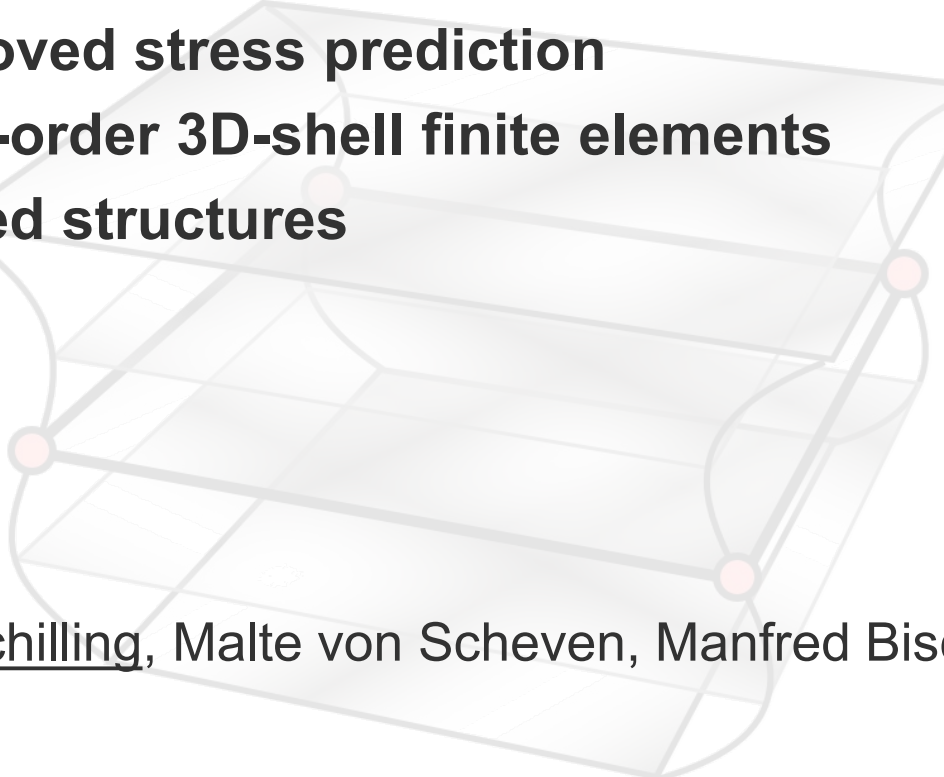


**University of Stuttgart**  
Institute for Structural Mechanics

**A comparative analysis  
of the improved stress prediction  
with higher-order 3D-shell finite elements  
for laminated structures**



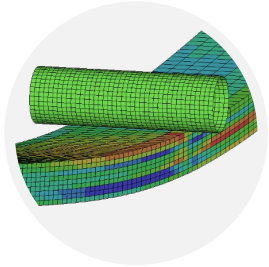
**Ansys Innovation  
Conference &  
17<sup>th</sup> LS-DYNA Forum**

October 16<sup>th</sup>, 2024  
in Leinfelden, Germany

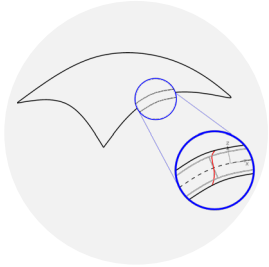
Maximilian Schilling, Malte von Scheven, Manfred Bischoff



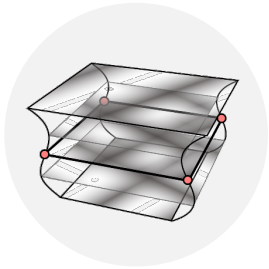
# Outline



**Simulation of laminated structures**



**Shell theories**



**Improved stress prediction with higher-order 3D-shell finite elements**



**Conclusion and outlook**

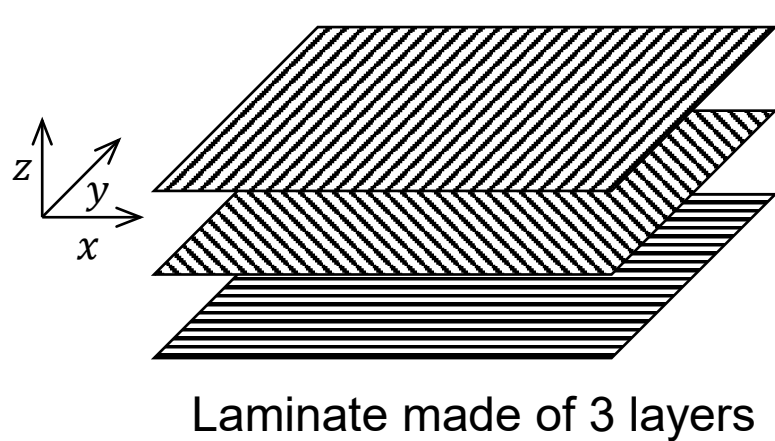
1

# Simulation of laminated structures

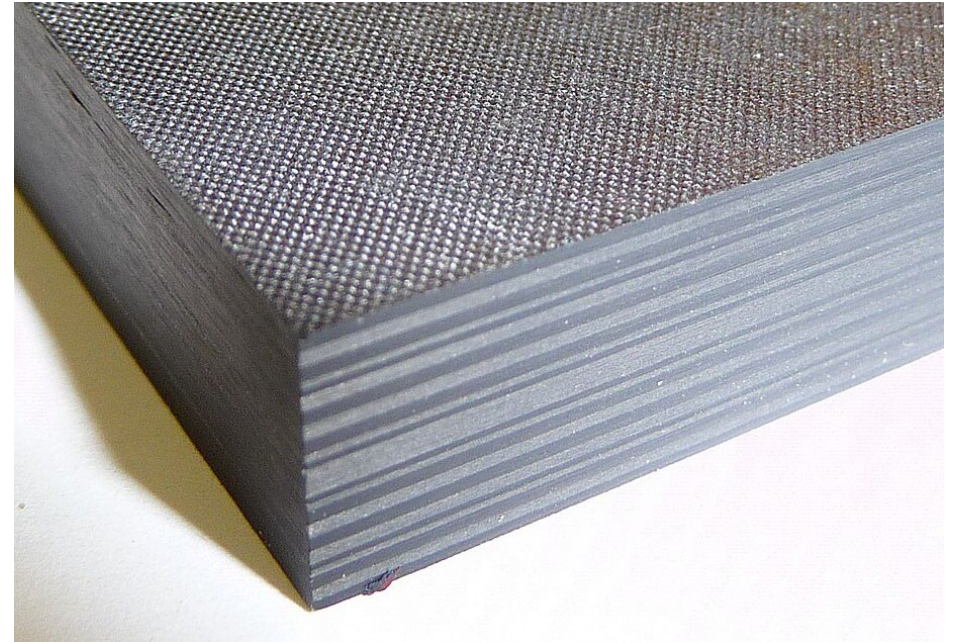
# Simulation of laminated structures

## Fundamentals of laminates

- Layers of fiber-reinforced material
- Orientation of fibers in each layer controls material properties
- Tailorable properties, high strength, and high stiffness of interest



90°  
-45°  
0°



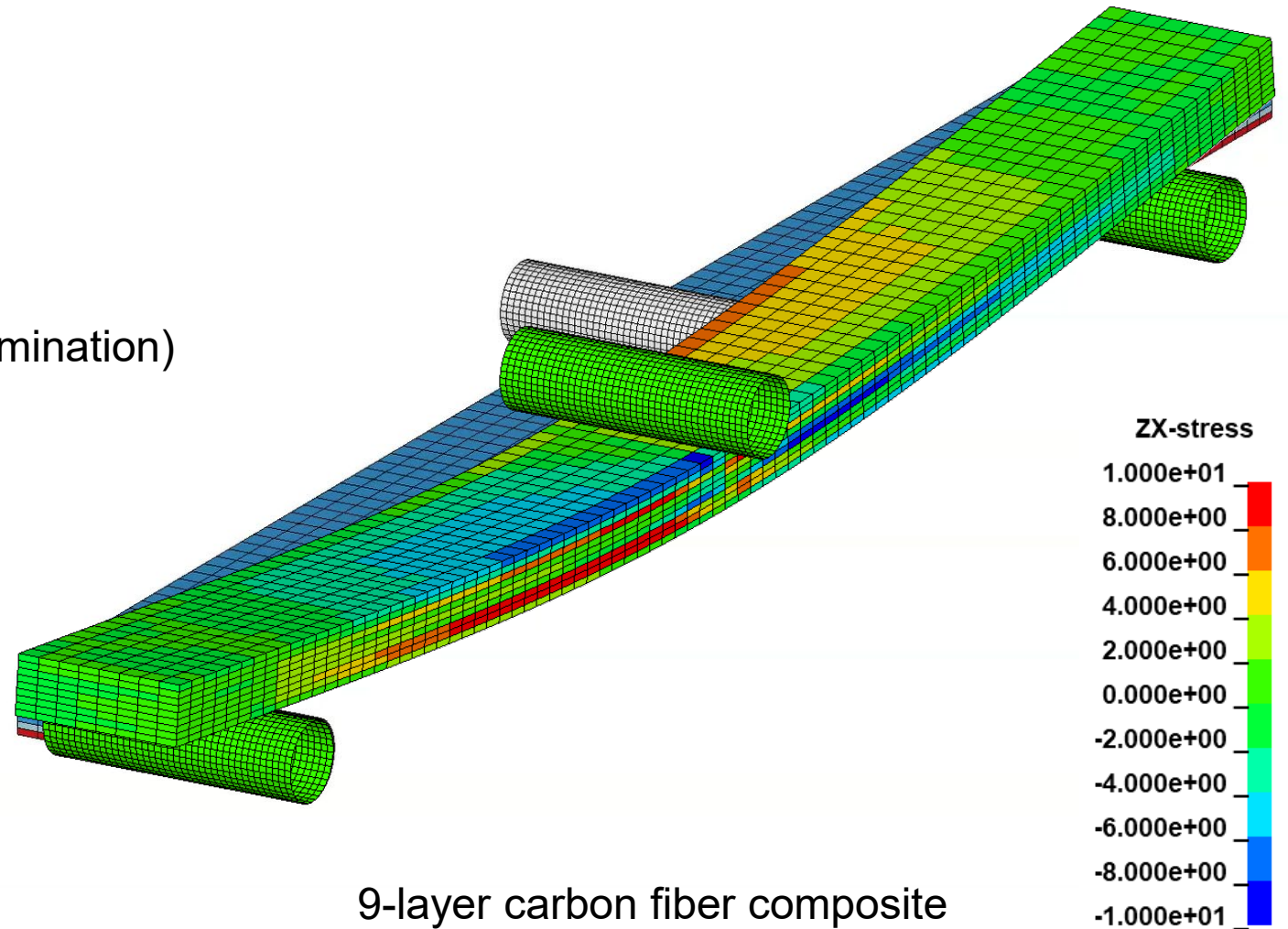
BY SIMON.WHITE.1000 - OWN WORK, CC BY-SA 3.0,  
[HTTPS://COMMONS.WIKIMEDIA.ORG/W/INDEX.PHP?CURID=19526520](https://commons.wikimedia.org/w/index.php?curid=19526520)

Fiber angle  $\beta$  relative to  $x$ -direction

# Simulation of laminated structures

## Challenges in finite element analysis of laminates

- Complex stress and strain fields
- Multiscale complexity
- Nonlinearities  
(e.g., intralaminar damage, delamination)



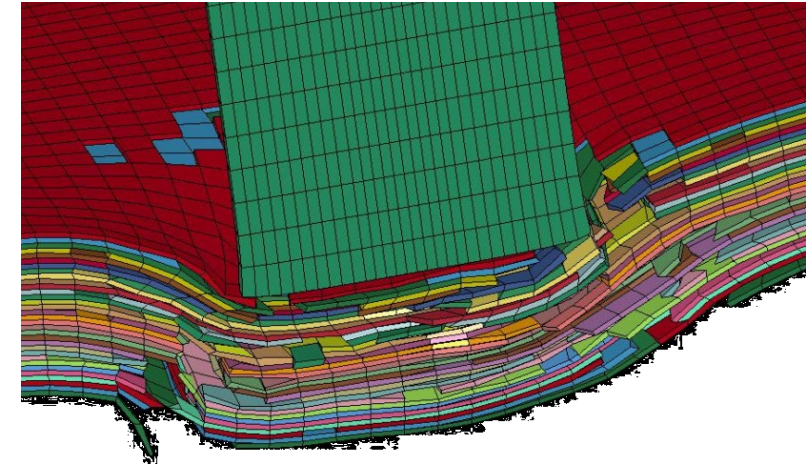
9-layer carbon fiber composite

# Simulation of laminated structures

## Current finite element modelling approaches

- **Meso scale: Components** (e.g. pressure vessel)

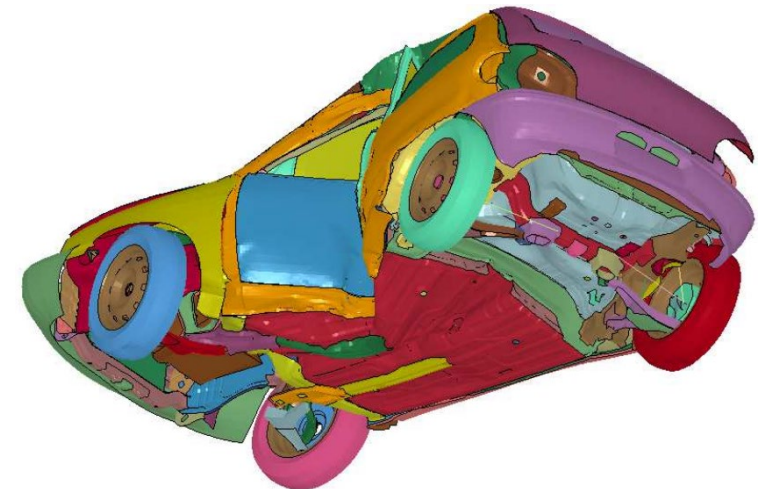
- Laminate as multiple layers of solid elements
- Complex, fully 3D stress state
- Higher accuracy



M.J. LOIKKANEN ET AL., SIMULATION OF BALLISTIC IMPACT ON COMPOSITE PANELS, 10TH LS-DYNA USERS CONFERENCE

- **Macro scale: Complete structures** (e.g., full vehicle crash)

- Laminate as a single layer of shell elements
- Reduced stress state
- Lower accuracy



E. NASSIOPOULOS ET AL., FINITE ELEMENT DYNAMIC SIMULATION OF WHOLE RALLYING CAR STRUCTURE: TOWARDS BETTER UNDERSTANDING OF STRUCTURAL DYNAMICS DURING SIDE IMPACTS, 8TH EUROPEAN LS-DYNA CONFERENCE  
PRESSURE VESSEL: [HTTPS://WWW.ASME.ORG/TOPICS-RESOURCES/CONTENT/FEA-ONLY-AS-GOOD-AS-THE-OPERATOR](https://www.asme.org/topics-resources/content/fea-only-as-good-as-the-operator)

# Simulation of laminated structures

## Current finite element modelling approaches

- **Meso scale: Components** (e.g. pressure vessel)
  - Laminate as a multiple layers of solid elements
  - Complex, fully 3D stress state
  - Higher accuracy

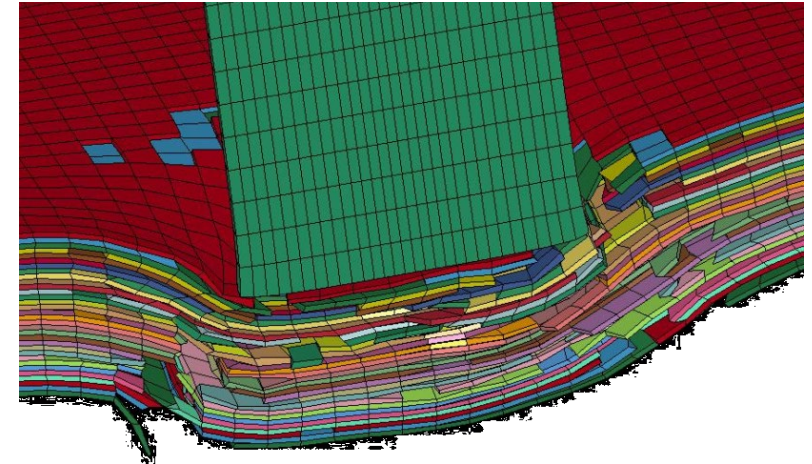
### What we would like:

→ Cheap, higher accuracy modelling approach for laminates in large-scale simulations.

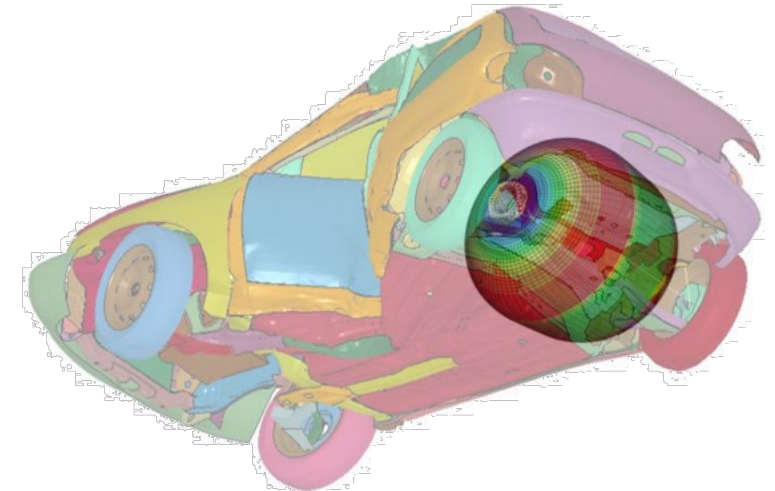
### One solution:

→ Higher-order 3D-shell elements

- **Macro scale: Complete structures** (e.g., full vehicle crash)
  - Laminate as a single layer of shell elements
  - Reduced stress state
  - Lower accuracy



M.J. LOIKKANEN ET AL., SIMULATION OF BALLISTIC IMPACT ON COMPOSITE PANELS, 10TH LS-DYNA USERS CONFERENCE



E. NASSIOPOULOS ET AL., FINITE ELEMENT DYNAMIC SIMULATION OF WHOLE RALLYING CAR STRUCTURE: TOWARDS BETTER UNDERSTANDING OF STRUCTURAL DYNAMICS DURING SIDE IMPACTS, 8TH EUROPEAN LS-DYNA CONFERENCE  
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2

# Shell theories



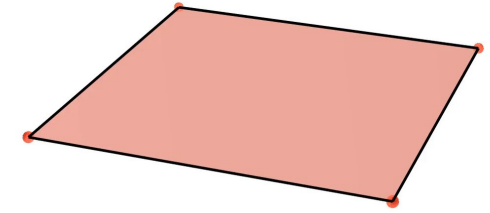
# Shell theories

## Reissner–Mindlin-shell (ELFORM={2, 16})

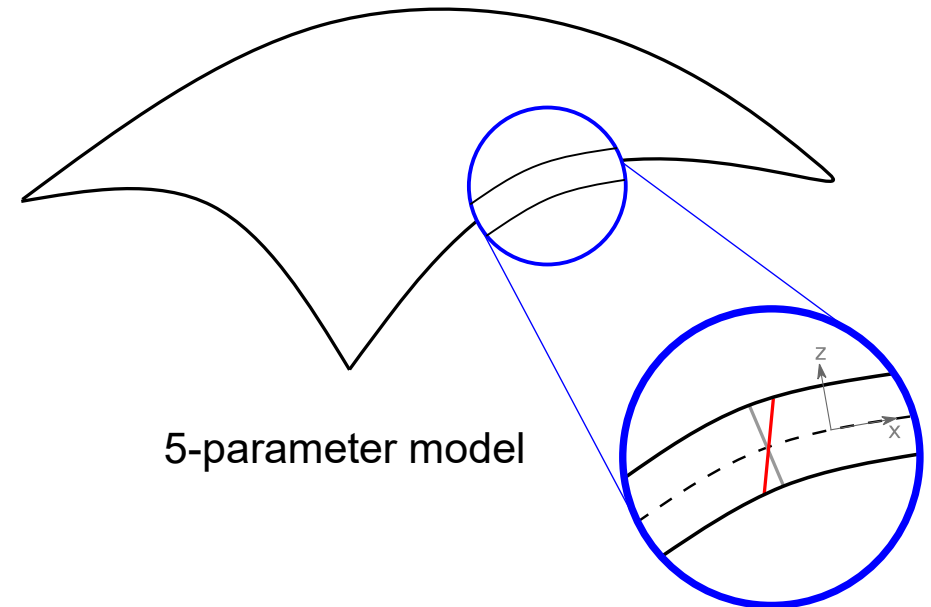
- Fibers transverse to the shell midface remain straight but not normal to the midface
- Transverse normal stress is zero

$$\boldsymbol{\sigma} = \begin{bmatrix} \sigma_{xx} & \sigma_{xy} & \sigma_{xz} \\ \sigma_{yx} & \sigma_{yy} & \sigma_{yz} \\ \sigma_{zx} & \sigma_{zy} & 0 \end{bmatrix}$$

- Modification of the material law



Reissner-Mindlin shell element



5-parameter model

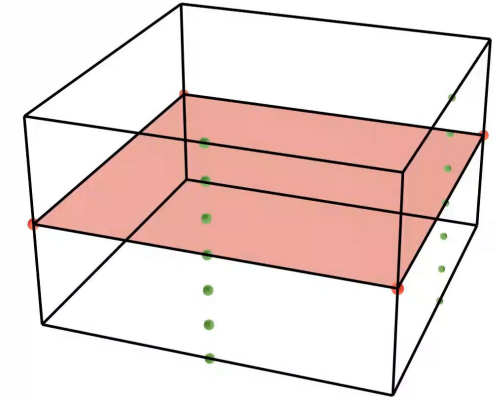
# Shell theories

## Shell with thickness stretch (ELFORM={25, 26})

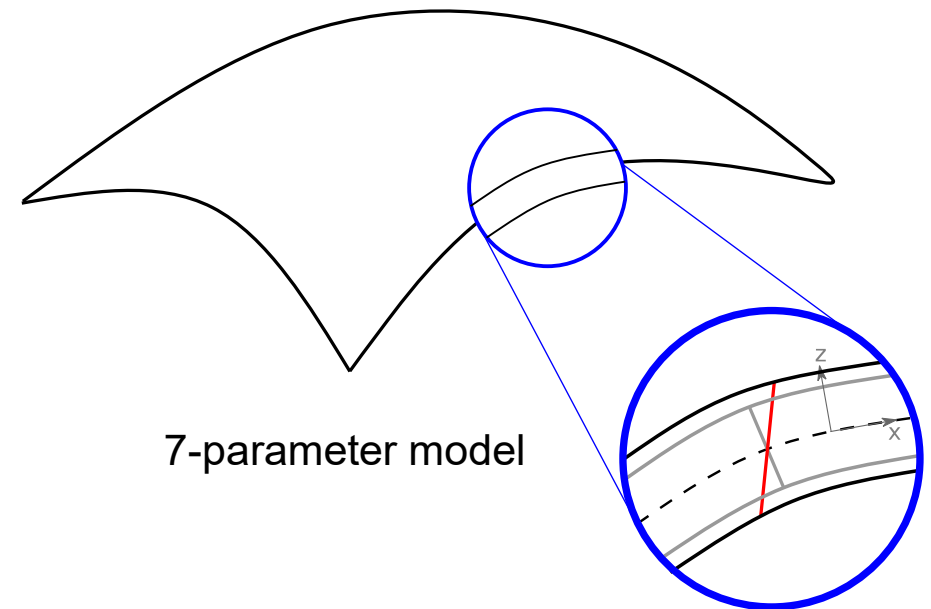
- Fibers transverse to the shell midface remain straight but not normal to the midface
- Transverse normal stress is **unequal to zero**
- **Thickness stretch and transverse normal strain**

$$\boldsymbol{\sigma} = \begin{bmatrix} \sigma_{xx} & \sigma_{xy} & \sigma_{xz} \\ \sigma_{yx} & \sigma_{yy} & \sigma_{yz} \\ \sigma_{zx} & \sigma_{zy} & \sigma_{zz} \end{bmatrix}$$

- **Fully three-dimensional material laws**



7p shell element



7-parameter model

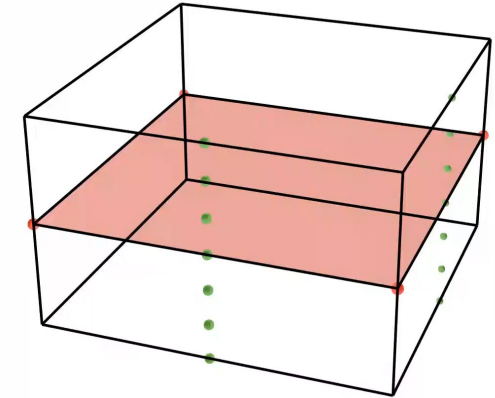
# Shell theories

## Higher-order 3D-shell

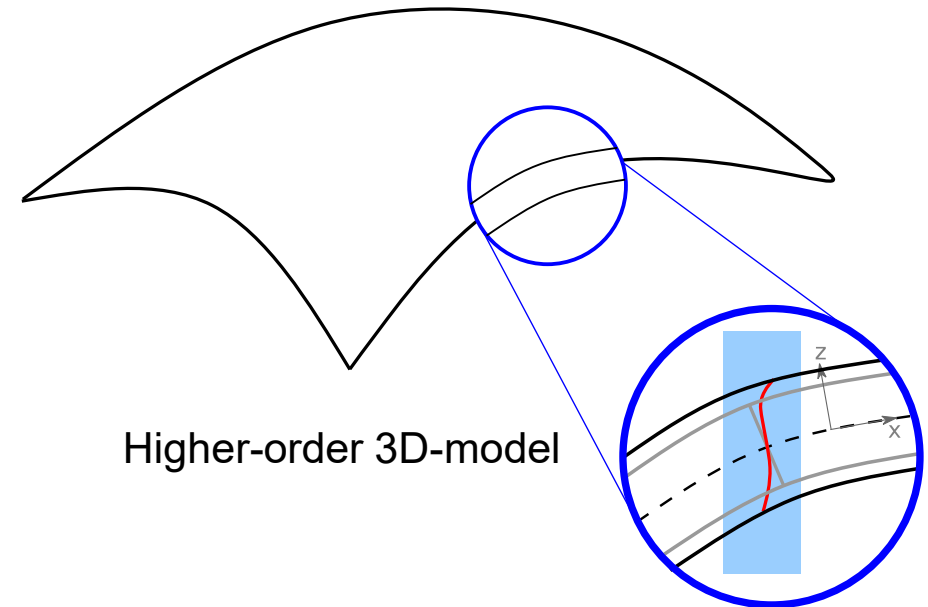
- Fibers transverse to the shell midface **do not remain straight** and not normal to the midface
- Transverse normal stress is unequal to zero
- **Nonlinear** thickness stretch and **nonlinear** transverse normal strain

$$\boldsymbol{\sigma} = \begin{bmatrix} \sigma_{xx} & \sigma_{xy} & \sigma_{xz} \\ \sigma_{yx} & \sigma_{yy} & \sigma_{yz} \\ \sigma_{zx} & \sigma_{zy} & \sigma_{zz} \end{bmatrix}$$

- Fully three-dimensional material laws
- Originally for sheet metal forming simulations
- Here: *Cubic 3D-shell element* (3DSH-cub) used



Higher-order 3D-shell element

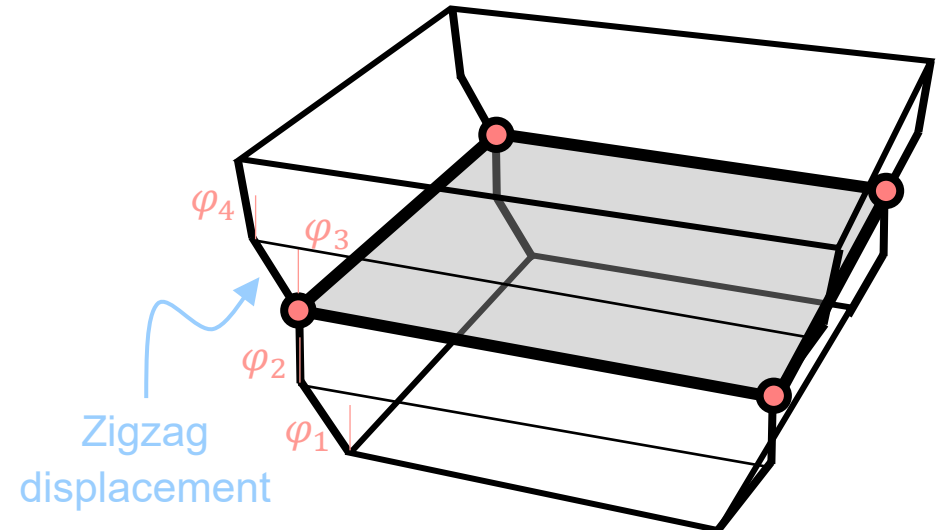


Higher-order 3D-model

## How to model laminates with shell finite elements

### 1. Kinematic layers

- Enhanced kinematics
- Additional degrees of freedom for each layer
- Computationally prohibitive

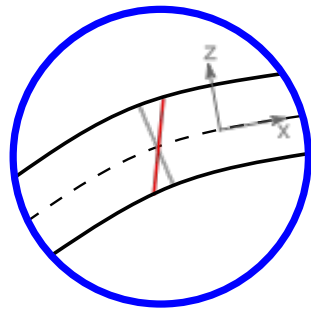
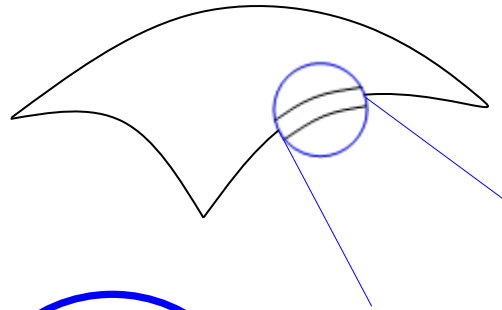


### 2. Numerical layers (quadrature points)

- Standard kinematics, no additional degrees of freedom
- $\geq 1$  quadrature point per layer
- Fiber angle  $\rightarrow$  material properties at quadrature point
- Computationally cheaper

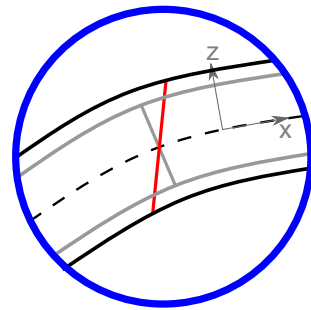
# Shell theories

## Overview



Reissner–Mindlin  
shell elements

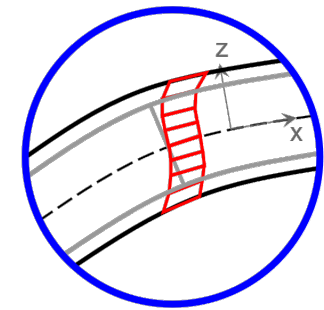
(ELFORM={2, 16})



Shell elements with  
thickness stretch

(ELFORM={25, 26})

**Higher-order  
3D-shell elements**



Solid elements

(ELFORM=-2)

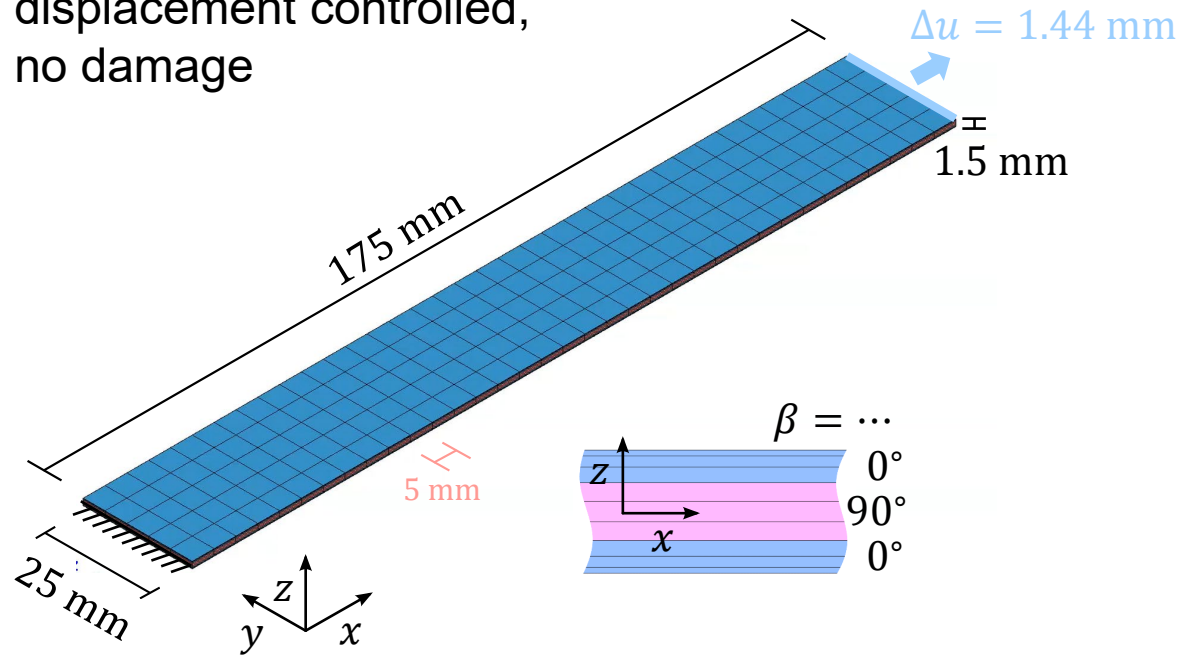
3

# **Improved stress prediction with higher-order 3D-shell finite elements**

# Improved stress prediction with cubic 3D-shell finite elements

## Tensile test (verification model)

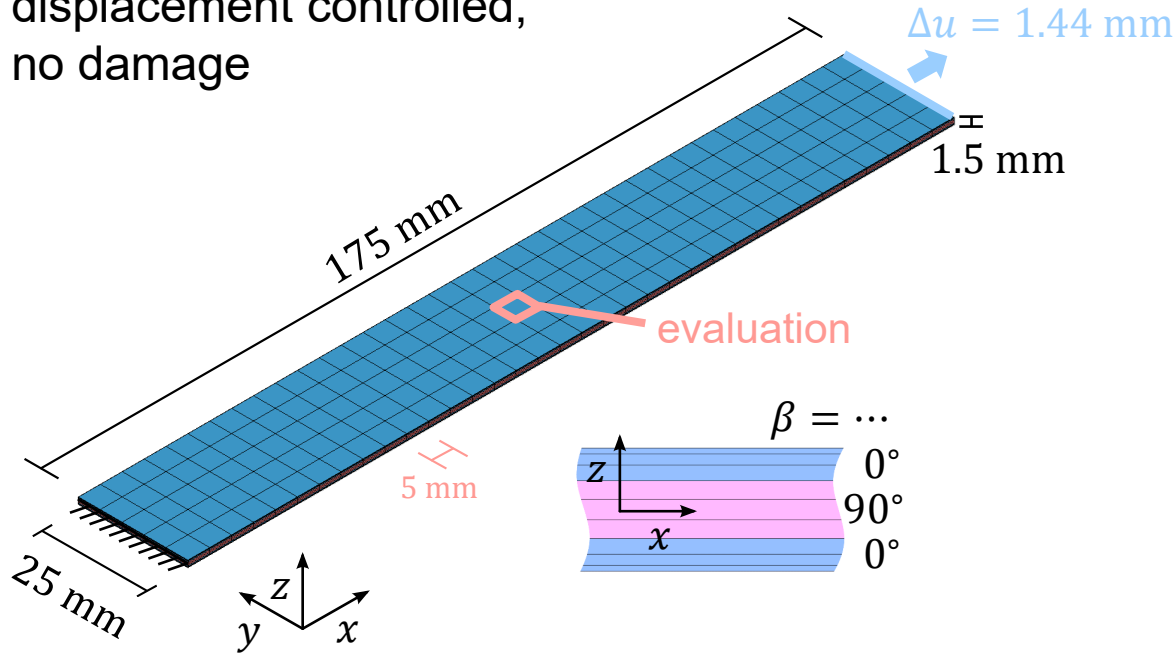
3-layer carbon fiber composite (\*MAT\_022),  
displacement controlled,  
no damage



# Improved stress prediction with cubic 3D-shell finite elements

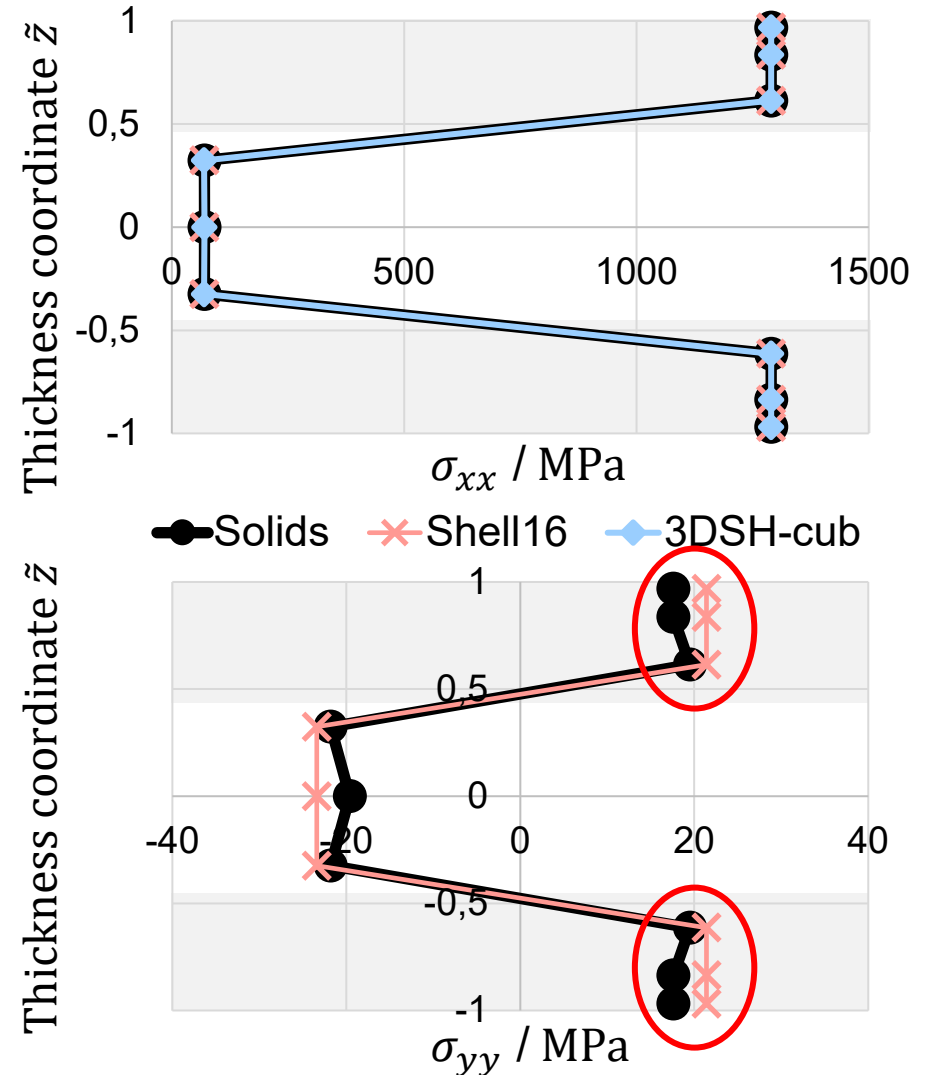
## Tensile test (verification model)

3-layer carbon fiber composite (\*MAT\_022), displacement controlled, no damage



Element formulations:

- 9 solids in thickness direction (reference) (ELFORM=-2)
- Reissner–Mindlin shell (ELFORM=16)
- Cubic 3D-shell (3DSH-cub)

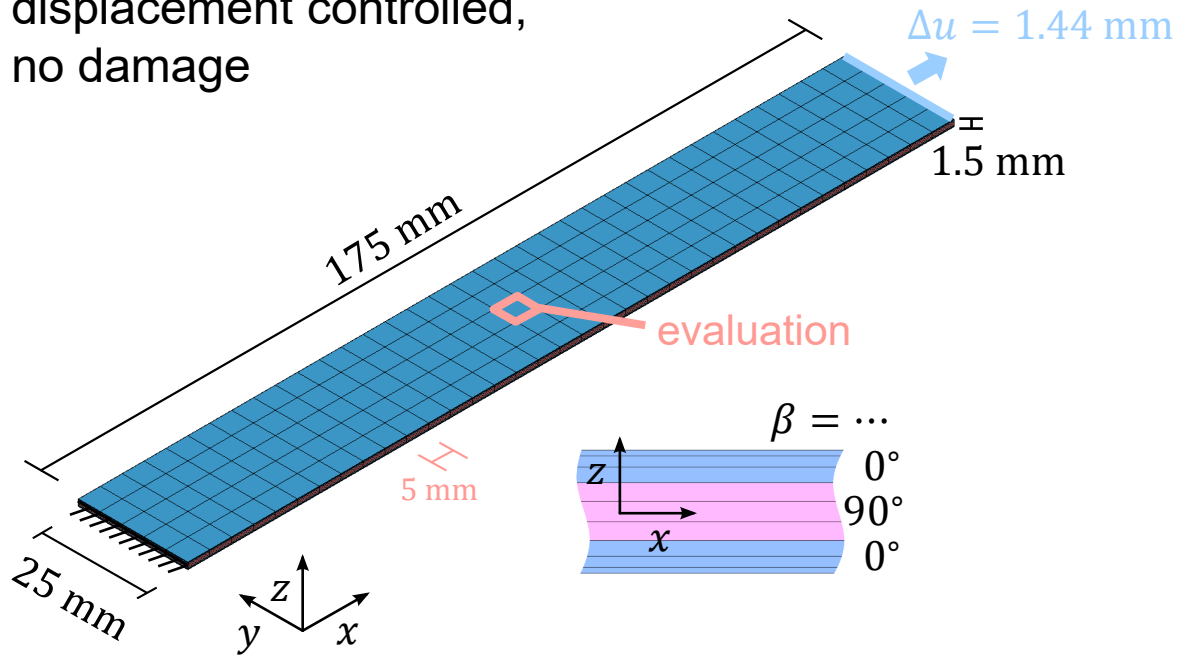




# Improved stress prediction with cubic 3D-shell finite elements

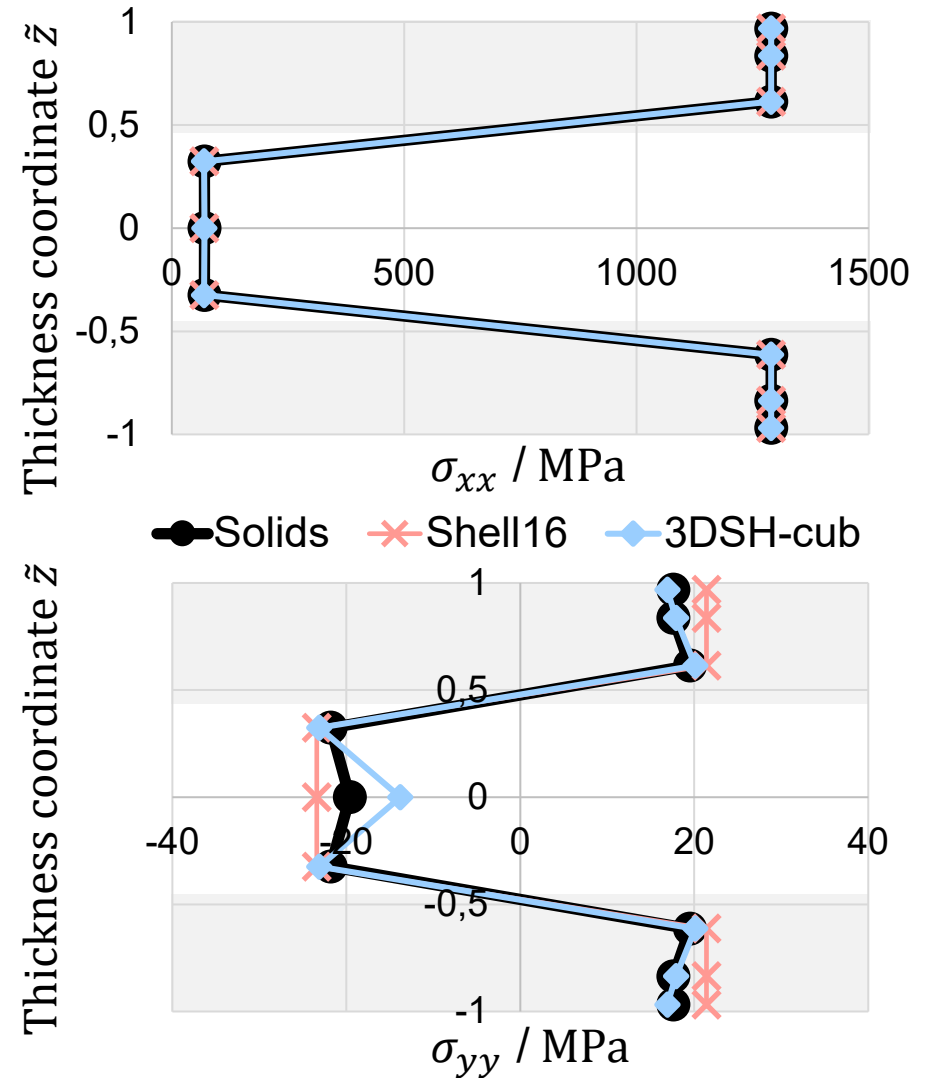
## Tensile test (verification model)

3-layer carbon fiber composite (\*MAT\_022), displacement controlled, no damage



Element formulations:

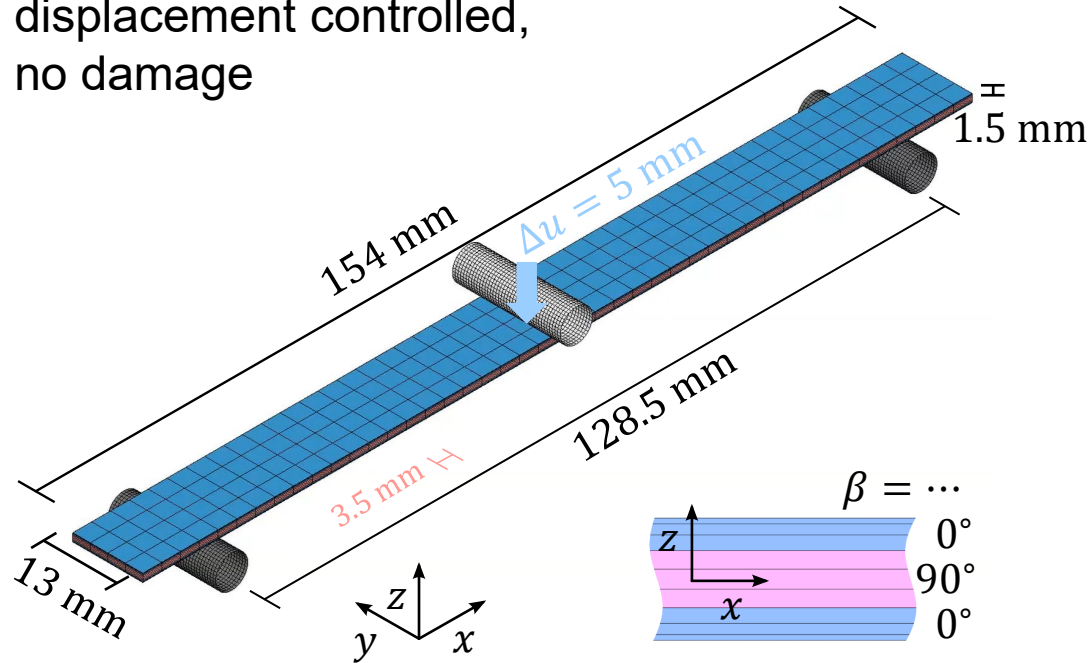
- 9 solids in thickness direction (reference) (ELFORM=-2)
- Reissner–Mindlin shell (ELFORM=16)
- Cubic 3D-shell (3DSH-cub)



# Improved stress prediction with cubic 3D-shell finite elements

## Three-point bending test (3 layers)

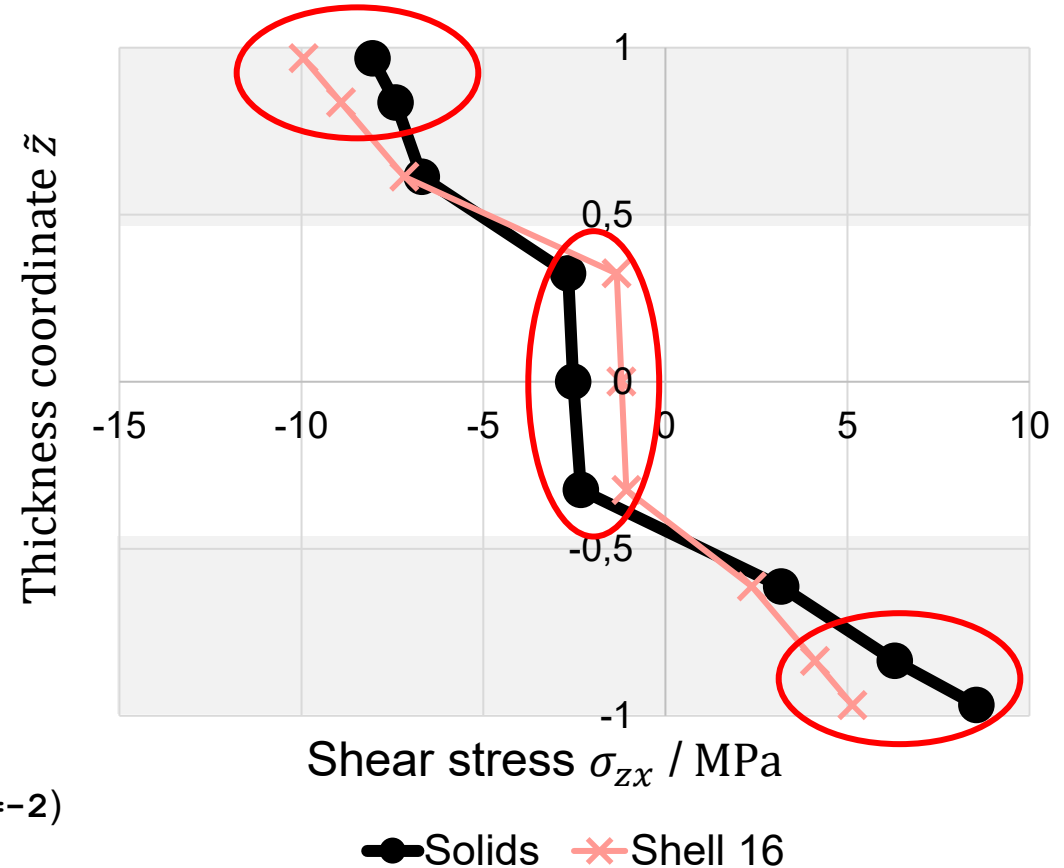
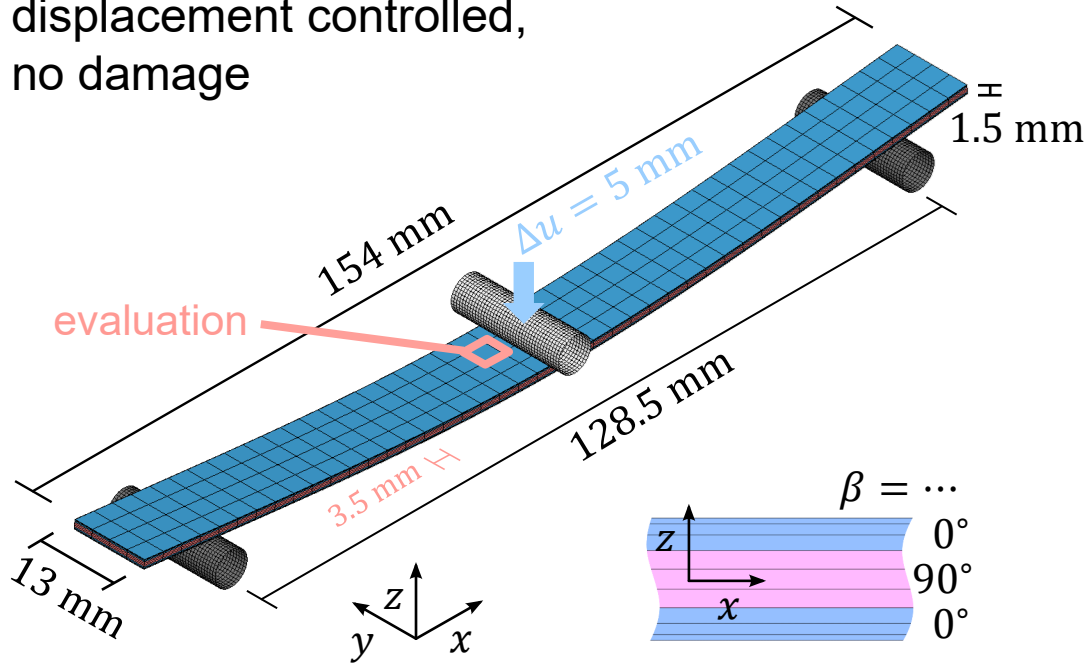
3-layer carbon fiber composite (\*MAT\_022),  
displacement controlled,  
no damage



# Improved stress prediction with cubic 3D-shell finite elements

## Three-point bending test (3 layers)

3-layer carbon fiber composite (\*MAT\_022),  
displacement controlled,  
no damage



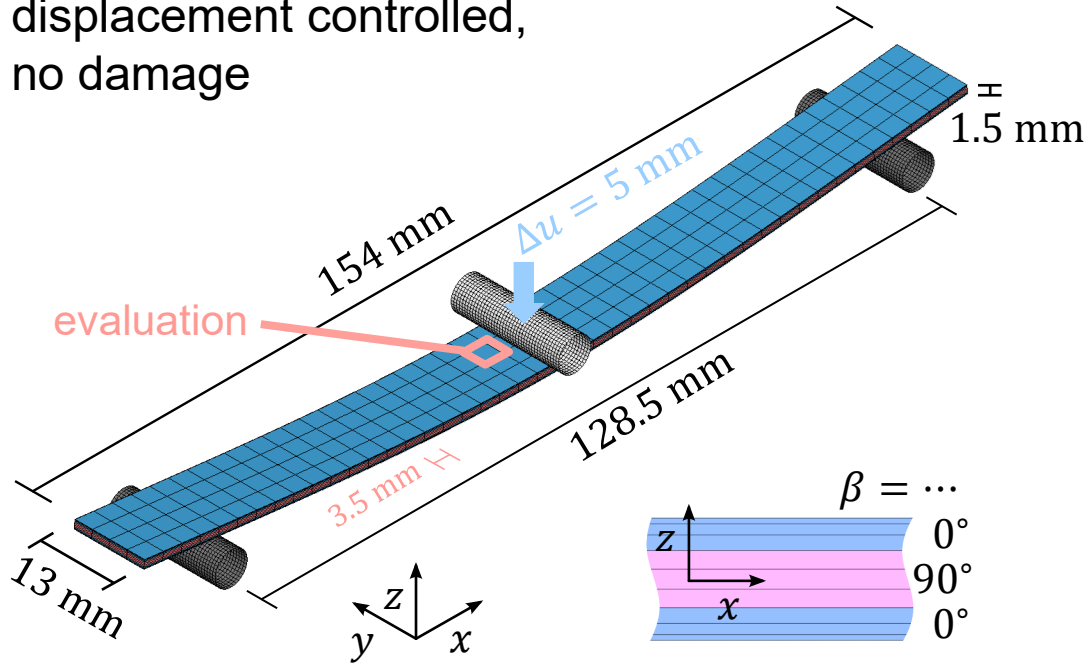
Element formulations:

- 9 solids in thickness direction (reference) (**ELFORM=-2**)
- Reissner–Mindlin shell (**ELFORM=16**)
- Cubic 3D-shell (**3DSH-cub**)

# Improved stress prediction with cubic 3D-shell finite elements

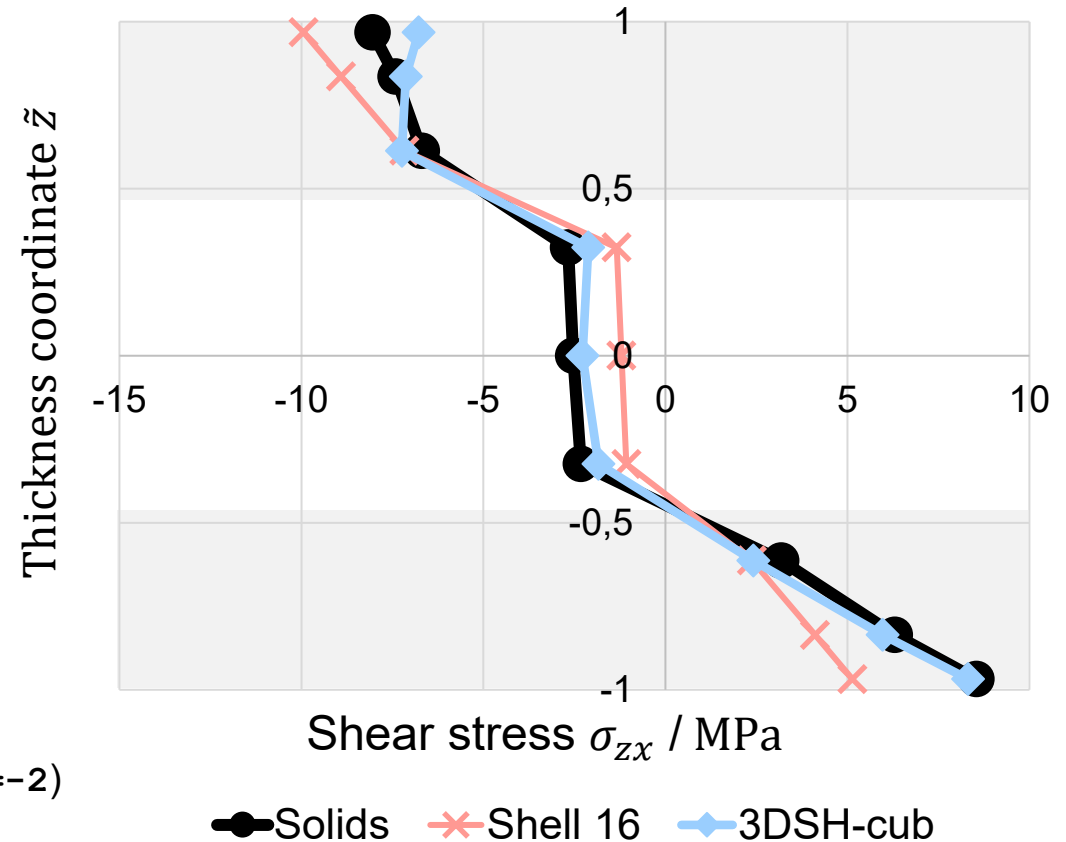
## Three-point bending test (3 layers)

3-layer carbon fiber composite (\*MAT\_022),  
displacement controlled,  
no damage



Element formulations:

- 9 solids in thickness direction (reference) (**ELFORM=-2**)
- Reissner–Mindlin shell (**ELFORM=16**)
- Cubic 3D-shell (**3DSH-cub**)

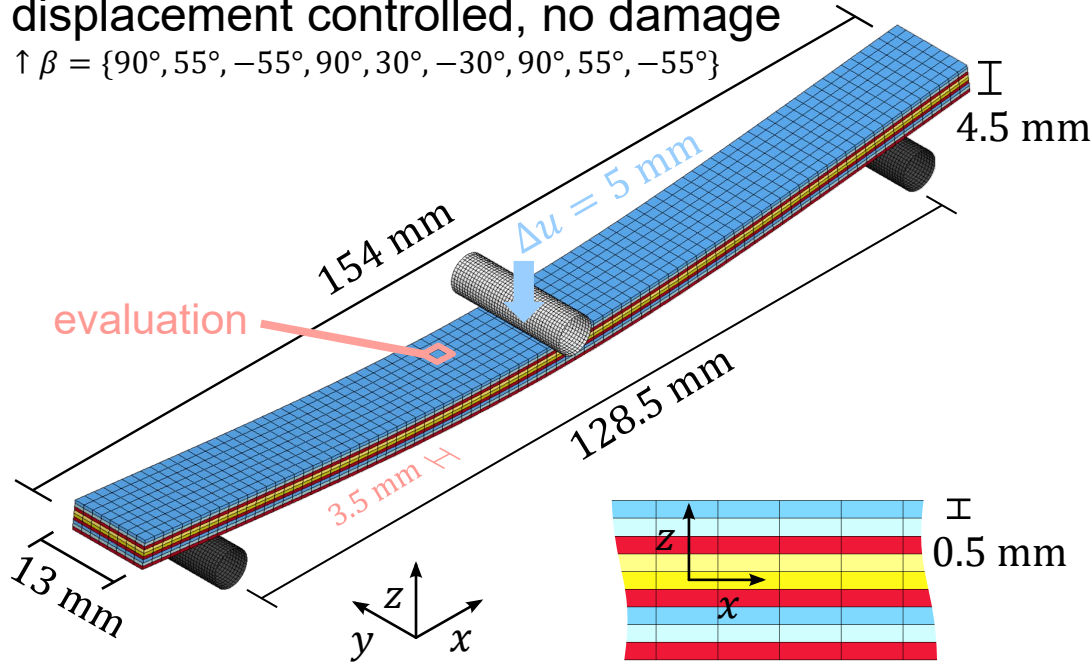


# Improved stress prediction with cubic 3D-shell finite elements

## Three-point bending test (9 layers)

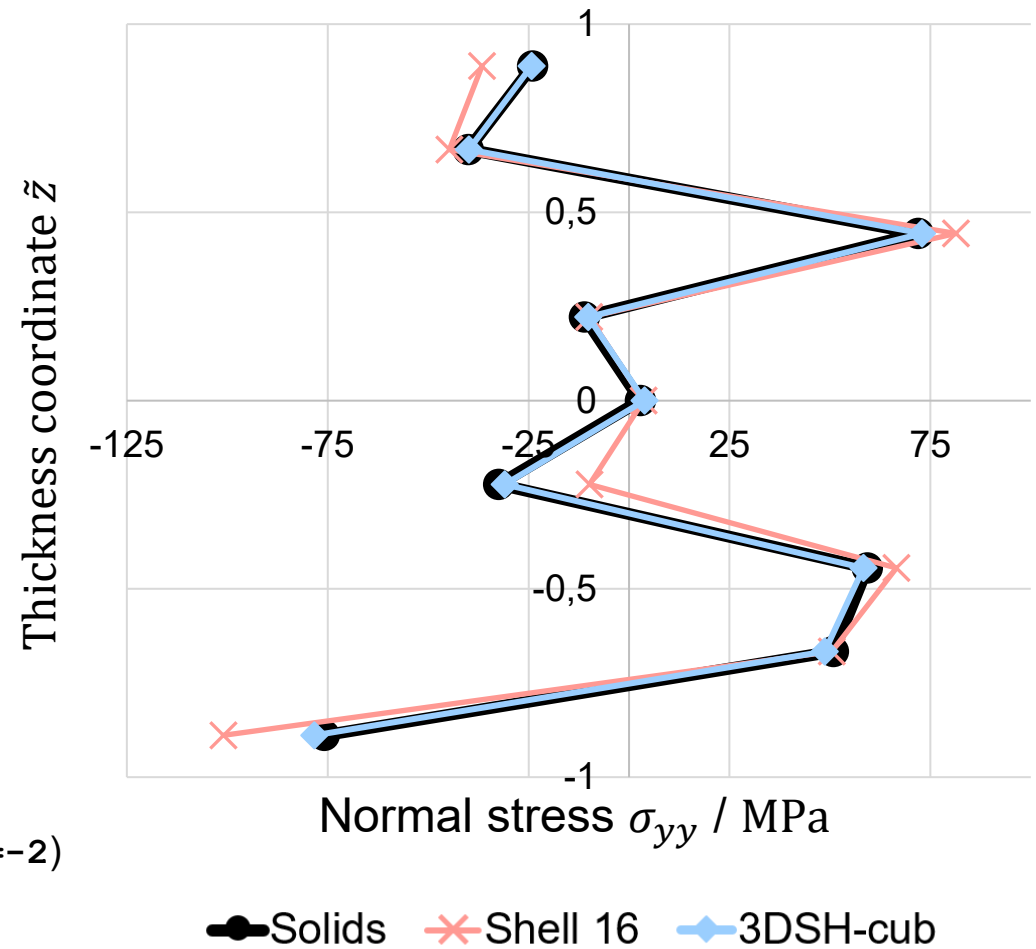
9-layer carbon fiber composite (\*MAT\_022),  
displacement controlled, no damage

$\uparrow \beta = \{90^\circ, 55^\circ, -55^\circ, 90^\circ, 30^\circ, -30^\circ, 90^\circ, 55^\circ, -55^\circ\}$



Element formulations:

- 9 solids in thickness direction (reference) (ELFORM=-2)
- Reissner–Mindlin shell (ELFORM=16)
- Cubic 3D-shell (3DSH-cub)

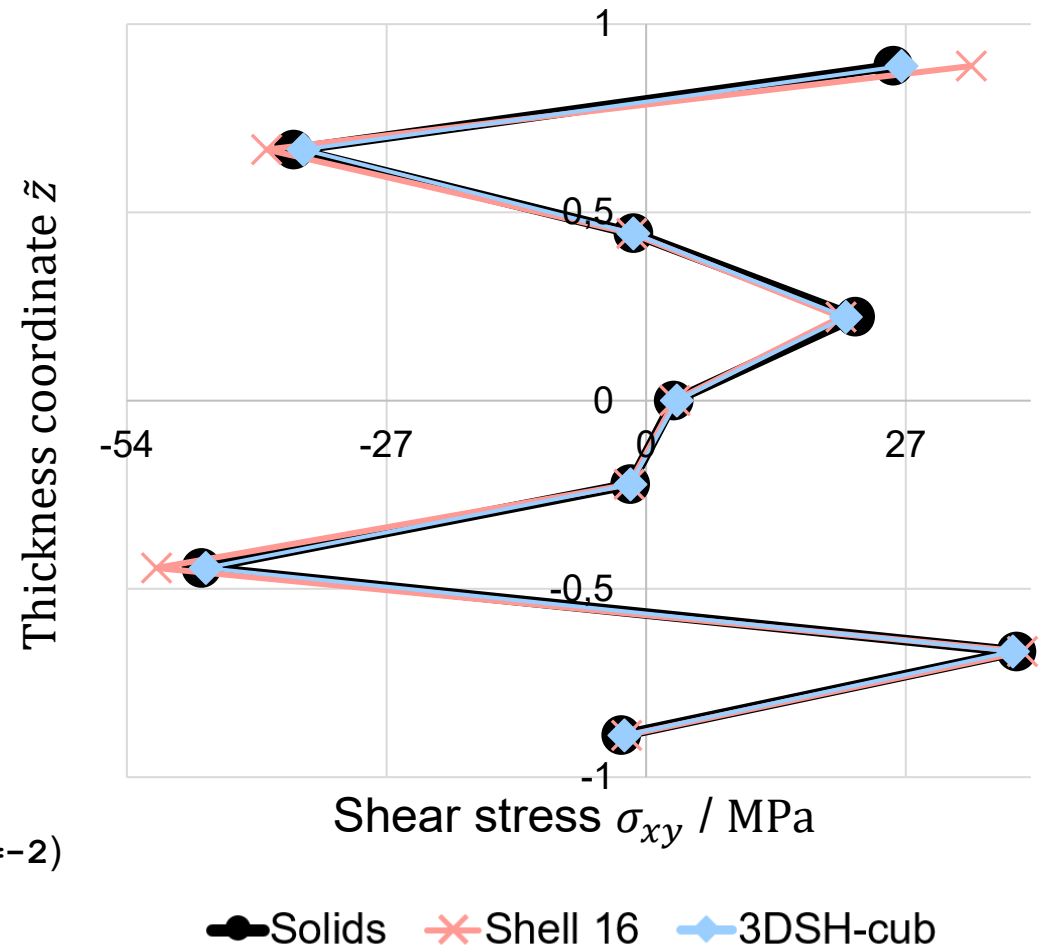
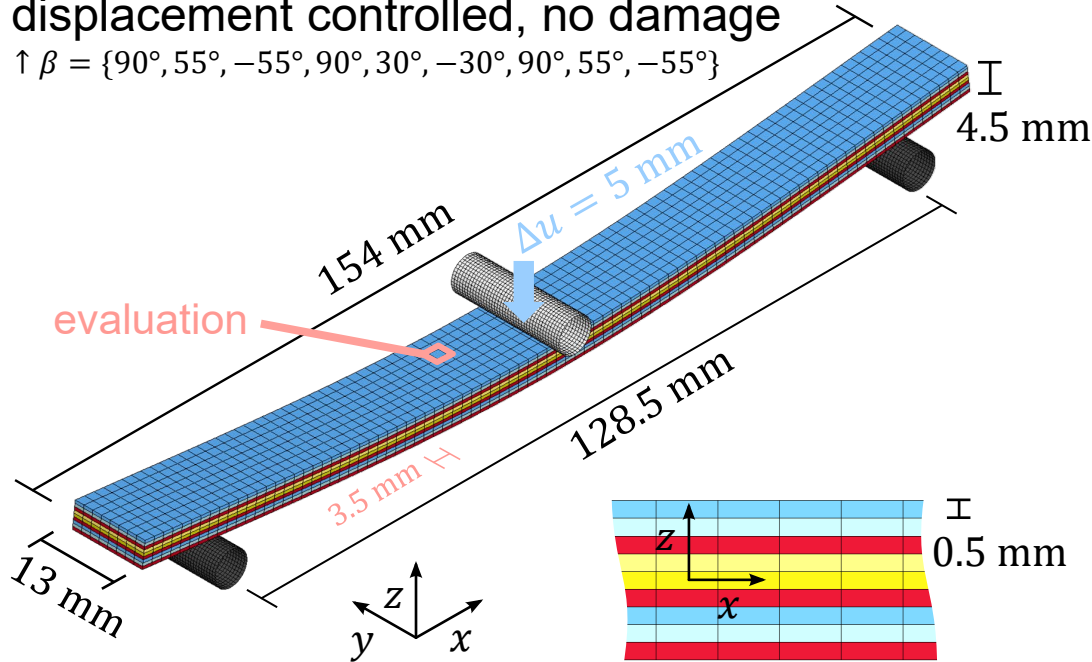


# Improved stress prediction with cubic 3D-shell finite elements

## Three-point bending test (9 layers)

9-layer carbon fiber composite (\*MAT\_022),  
displacement controlled, no damage

$\uparrow \beta = \{90^\circ, 55^\circ, -55^\circ, 90^\circ, 30^\circ, -30^\circ, 90^\circ, 55^\circ, -55^\circ\}$



Element formulations:

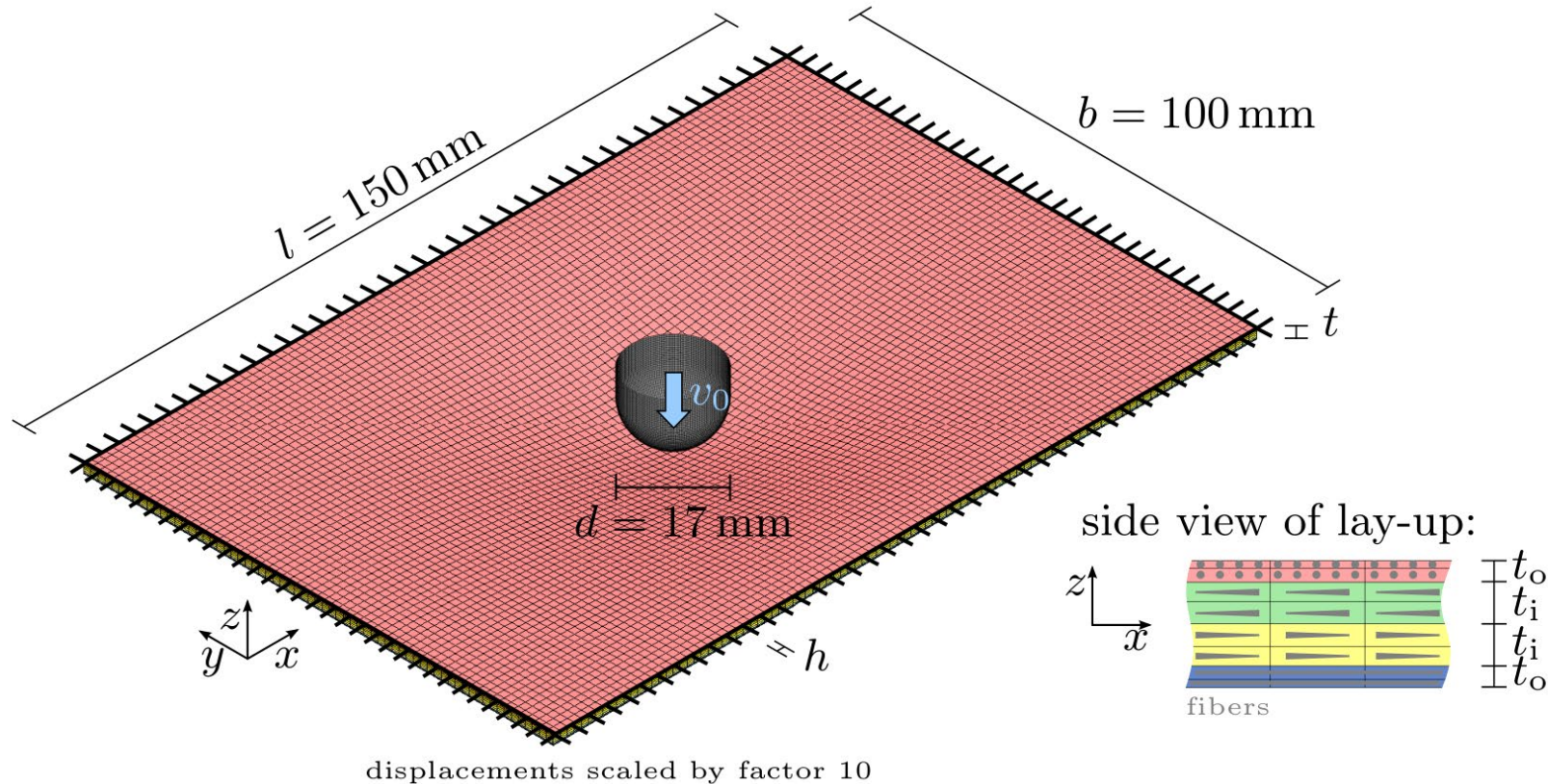
- 9 solids in thickness direction (reference) (ELFORM=-2)
- Reissner–Mindlin shell (ELFORM=16)
- Cubic 3D-shell (3DSH-cub)

# Improved stress prediction with cubic 3D-shell finite elements

## Impact test

4-layer carbon fiber composite (\*MAT\_022),  
impactor with initial velocity  $v_0$  (slow impact)

$$\uparrow \beta = \{0^\circ, 45^\circ, -45^\circ, 90^\circ\}$$

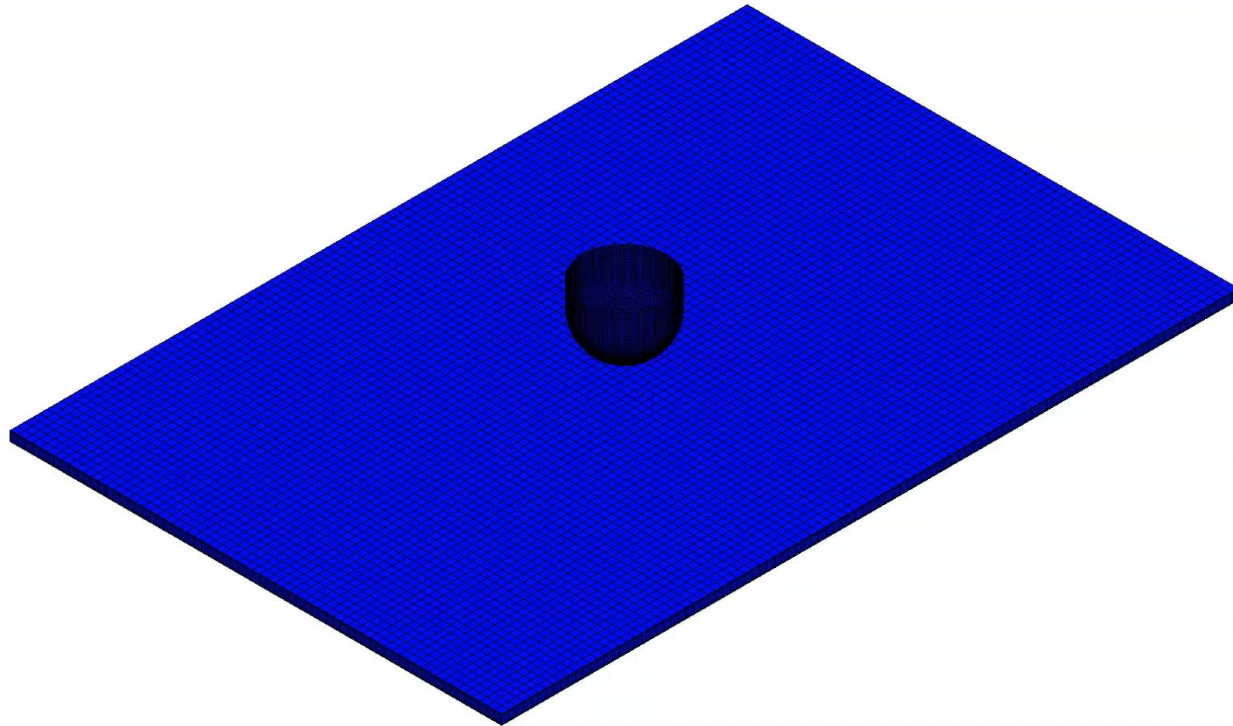


# Improved stress prediction with cubic 3D-shell finite elements

## Impact test

4-layer carbon fiber composite (\*MAT\_022),  
impactor with initial velocity  $v_0$  (slow impact)

$\uparrow \beta = \{0^\circ, 45^\circ, -45^\circ, 90^\circ\}$



displacements scaled by factor 10

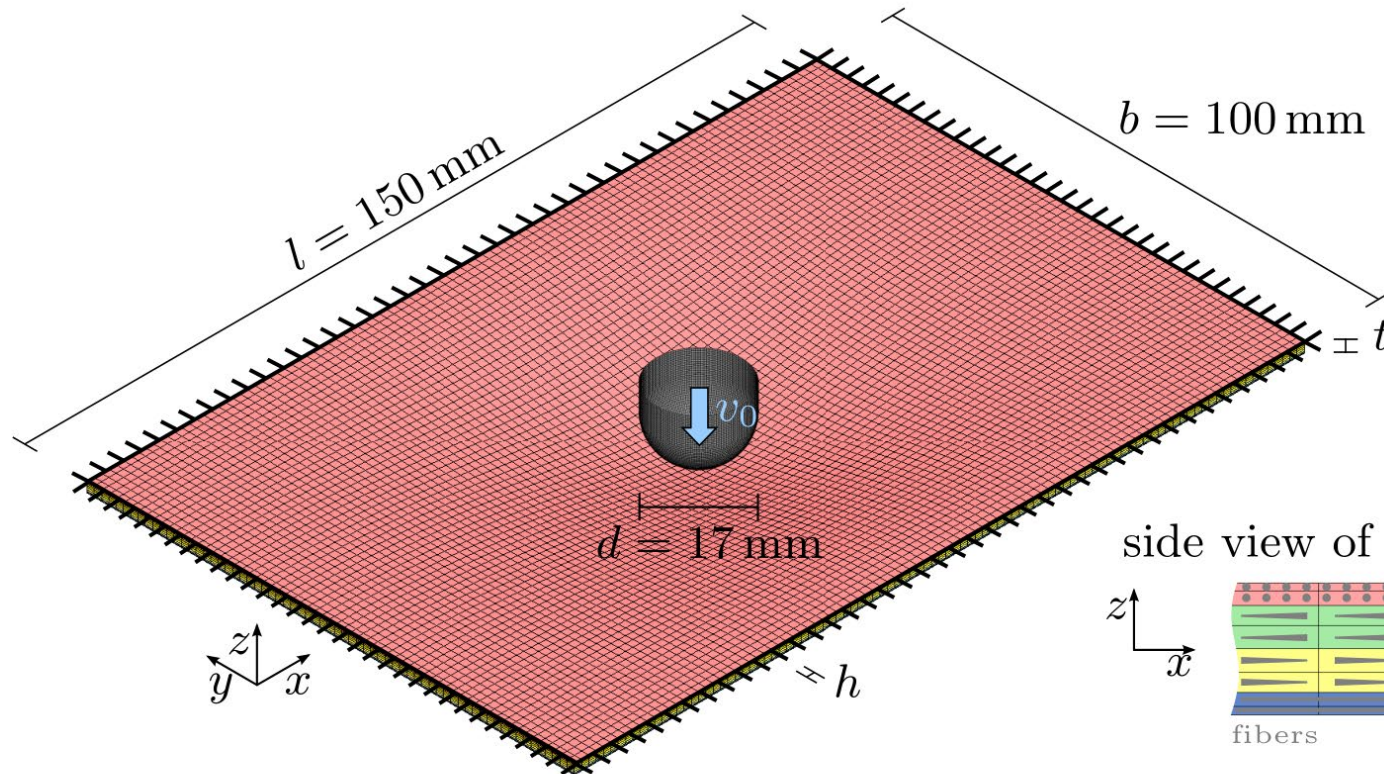


# Improved stress prediction with cubic 3D-shell finite elements

## Impact test

4-layer carbon fiber composite (\*MAT\_022),  
impactor with initial velocity  $v_0$  (slow impact)

$$\uparrow \beta = \{0^\circ, 45^\circ, -45^\circ, 90^\circ\}$$

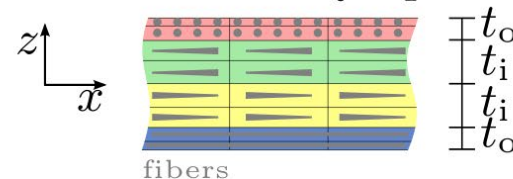


displacements scaled by factor 10

Element formulations:

- 8 solids in thickness direction (reference) (ELFORM=-2)
- Reissner–Mindlin shells (ELFORM=16)
- Conventional 3D-shells (ELFORM=26)
- Cubic 3D-shells (3DSH-cub)

side view of lay-up:



# Improved stress prediction with cubic 3D-shell finite elements

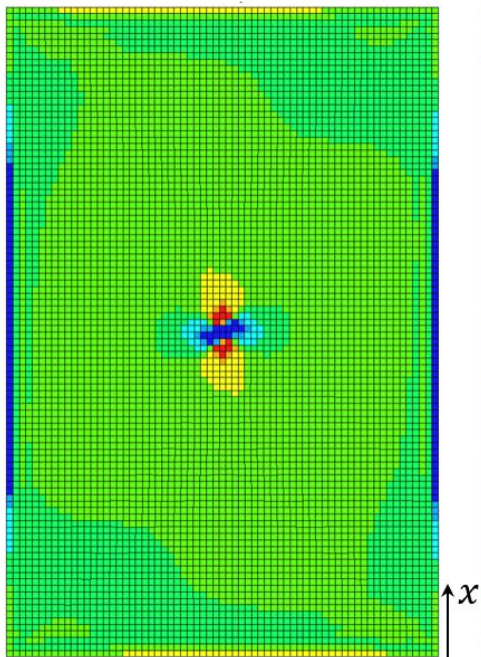
## Impact test

4-layer carbon fiber composite (\*MAT\_022),  
impactor with initial velocity  $v_0$  (slow impact)

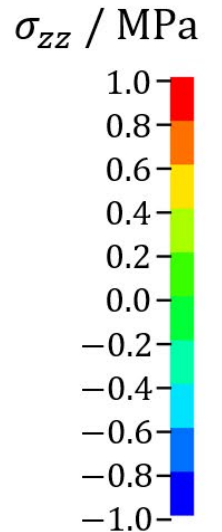
$$\uparrow \beta = \{0^\circ, 45^\circ, -45^\circ, 90^\circ\}$$

*Only shell element to  
qualitatively align  
with reference solution*

Contour plot of normal stress  $\sigma_{zz}$  in 45° layer



8 solids (ref.)



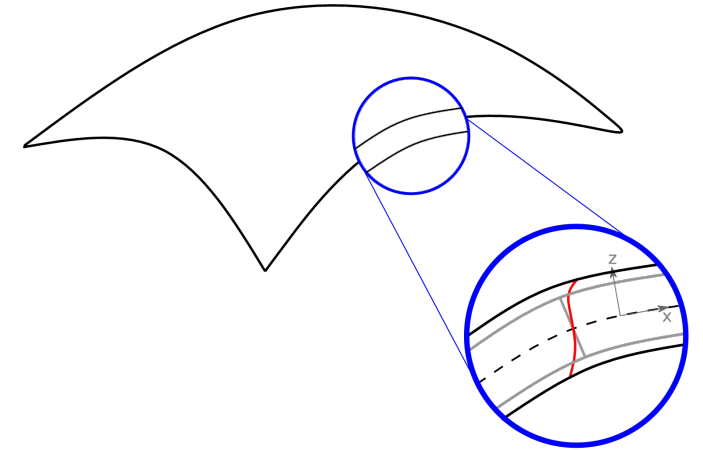
4

# Conclusion and outlook



## Higher-order 3D-shell elements...

- ... can offer an improved prediction quality for stress in laminated structures.
- ... enable large scale simulations with a 3D material law, taking into account all stress components.



Industrial scale examples,  
detailed analysis of impact scenarios,  
enhancements for delamination.

## Acknowledgement

This research has been funded by the project DigiTain 19S22006K by the Federal Ministry of Economic Affairs and Climate Action based on a resolution of the German Bundestag. This support is gratefully acknowledged.



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the European Union  
NextGenerationEU

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on the basis of a decision  
by the German Bundestag



Universität Stuttgart

Thank you!



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Baustatik und Baudynamik (IBB)



@uni\_stuttgart\_ibb

